Amendments to the Specification:

Please replace the paragraphs numbered [0014] with the following:

The first passages 66 are in fluid communication with a second portion 74.2 of a second cavity 74 that is bounded by a portion of the second side 72 of the rotor 24 and by a second bounding surface of a forward cover 5076, wherein the forward cover 50-76 comprises an intermediate rim 78 and an outer rim 80 that engage respective first 82.1 and second 82.2 lips formed on the second side 72 of the rotor 24. The outer rim 80 is sealed to the second lip 82.2 so as to prevent leakage of fuel 12 from the joint therebetween. The intermediate rim 78 incorporates at least one passage 84 that provides for fluid communication between the second portion 74.2 of the second cavity 74 and a first portion 74.1 thereof. The first portion 74.1 of the second cavity 74 is in fluid communication with the interior 86 of a shaft 88 of the shaft assembly 28 via at least one passage 90 through the shaft 88, and the interior 86 of the shaft 88 is in fluid communication with a first discharge orifice 92 through at least one other passage 94 through the shaft 88. The first discharge orifice 92 is in fluid communication with the combustion chamber 16, and thereby provides for a discharge of fuel 12 directly from the rotating shaft 88 to the combustion chamber 16. The first discharge orifice 92 is, for example, a part of a second rotary fluid trap 96 that provides for isolating the relatively high pressure of the combustion chamber 16 from the relatively lower pressure of the interior of the shaft 88 and the first portion 74.1 of the second cavity 74, whereby the principles of structure and operation of the second rotary fluid trap 96 are the same as those of the first rotary fluid trap 42 described hereinabove.

Application No. 10/709,199 Amendment Dated 23 July 2005 Response to Office Action Dated 24 June 2005

Please replace the paragraphs numbered [0016] with the following:

Accordingly, the gas turbine engine 10 comprises a rotatable portion 118 that is rotatable with respect to a housing 34 of the gas turbine engine 10, wherein the rotatable portion 118 comprises the turbine 20 / bladed rotor 22, comprising the rotor 24 and the blades 26; the aft cover 50 and associated first rotary fluid trap 42; the forward cover 5076; and the shaft assembly 28 / shaft 88 and associated first discharge orifice 92 / second rotary fluid trap 96, all of which rotate in unison with a rotating frame of reference. After discharge from the relatively fixed orifice 38, the fuel 12 is contained within the rotatable portion 118 until discharge directly into the combustion chamber 16 from the first discharge orifice 92 of the rotatable portion 118 in the rotating frame of reference Accordingly, because all of the elements of the rotatable portion 118 rotate in unison with the rotating frame of reference, these elements can be readily sealed to one another as necessary to contain the fuel 12 therein, for example, at the junctions of the outer rims 60, 80 of the first 50 and second 76 bounding surfaces with the second lips 62.2, 82.2 of the rotor 24, which could otherwise be problematic if it were necessary to provide for sealing across a relatively moving junction of elements to be sealed to one another.